NASM1312-11 29 August 1997

ADOPTION NOTICE

NASM1312-11, "Fastener Test Methods Method 11 Tension Fatigue" was adopted on 29 August 1997 for use by the Department of Defense (DoD). Proposed changes by DoD activities must be submitted to the DoD Adopting Activity: Commander, Naval Air Warfare Center Aircraft Division, Code 414100B120-3 Highway 547, Lakehurst, NJ 08733-5100. DoD activities may obtain copies of this standard from the Standardization Document Order Desk, 700 Robbins Avenue, Building 4D, Philadelphia, PA 19111-5094. The private sector and other Government agencies may purchase copies from the Aerospace Industries Association, 1250 Eye Street NW, Washington, DC 20005.

NASM1312-11 Should be used instead of MIL-STD-1312-11A, which was cancelled on 29 August 1997.

Custodians:

Army - AV

Navy - AS

Air Force - 11

Review Activities:

Army - AV, AR

Navy - AS, SH

Air Force - 11

DLA - IS

Preparing activity: Navy - AS (Project No. 53GP-0285-11)

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

> NASM1312-11 STANDARD PRACTICE

FASTENER TEST METHODS

METHOD 11

TENSION FATIGUE



THE INITIAL RELEASE OF THIS DOCUMENT SUPERSEDES MIL-STD-1312-11A

DESIGNATION FOR THIS TEST METHOD REMAINS MIL-STD-1312-11

LIST OF CURRENT SHEETS									
NO. 1 2 3 4 5 6 7 8 9									9
REV.	NEW								
NO.	10	11	12	13	14	15			
REV.	NEW	NEW	NEW	NEW	NEW	NEW			

FSC 53GP

SHEET 1 0F 15

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC.
ALL RIGHTS RESERVED

NASM1312-11

FOREWORD

This standard sets forth standard test procedures for determining tension fatigue of bolts, screws and nuts at room temperature.

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

M	CA	113	10	1	1
INF	1.71	α		- 1	- 1

	TABLE OF CONTENTS	11.	
Paragraph	TABLE OF CONTENTS	الماء الموا	Sheet
1.	SCOPE		-
1.1			
1.1	Applicability	•••••	5
2.	REFERENCED DOCUMENTS	•••••	5
2.1	Government documents		
2.1.1	Specifications, standards and handbooks		
2.2	Other publications		
3.	DEFINITIONS		6
4.	GENERAL REQUIREMENTS		6
4.1	Test apparatus		
4.1.1	Fixtures		
4.1.2	Static alignment		
4.1.2.1	Spacer		
4.1.2.2	Test nuts		6
4.1.2.3	Test bolts		
4.1.2.4	Limitations		
4.1.2.5	Fixture Alignment		
4.1.2.6	Threaded adapters		10
4.2	Test specimens		10
4.2.1	Bolts		
4.2.2	Testing of nuts		
4.2.3	Nut test results		
5.	DETAIL REQUIREMENTS		10
5.1	Test procedures		
5.1.1			
5.1.1.1	Bolts Tension bolts		10
5.1.1.2	Shear bolts	- N:41Å	10
5.1.1.3	Flush head bolts		11
5.2	Test conditions.		
5.2.1	Torque		
5.2.2	Load level		
5.2.3	Load ratio		
5.2.4	Test speed		
6.	NOTES		11
6.1	Test report		

SHEET 3

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11

FIGURES

9 1 10 1

Figure Property of the Figure 1		Sheet
1.	Typical test fixture arrangement	13
2	Nut, fatigue test, room temperature, for tension bolts	14
3.	Nut, fatigue test, room temperature, for shear bolts	15
	TABLES	
<u>Table</u>		Sheet
I.	List of items called out on figure 1	7
II.	List of dimensions for figure 2	8
III.	List of dimensions for figure 3	

COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11

- 1. SCOPE
- 1.1 Applicability. This test method establishes the procedure and equipment to be used for room temperature tension fatigue testing of bolts, screws and nuts.
- 2. REFERENCED DOCUMENTS
- 2.1 Government documents.
- 2.1.1 <u>Specifications, standards and handbooks</u>. Unless otherwise specified, the following specifications, standards and handbooks of the issue listed in the current Department of Defense Index of Specifications and Standards (DoDISS) and the supplement thereto (if applicable), form a part of this standard to the extent specified herein.

SPECIFICATIONS

MILITARY

MIL-I-6868 Inspection Process, Magnetic Particle
 MIL-H-6875 Heat Treatment of Steels (Aircraft Practice), Process for
 MIL-S-8879 Screw Threads, Controlled Radius Root with
 Increased Minor Diameter, General Specification for
 MIL-C-15074 Corrosion Preventive, Fingerprint Remover

STANDARDS

MILITARY

MIL-STD-1312 Fastener Test Methods

(Copies of specifications, standards, handbooks, drawings and publications required by contractors in connection with specific acquisition functions should be obtained from the contracting activity, or as directed by the contracting officer.)

2.2 Other publications. The following document(s) forms a part of this specification to the extent specified herein. The issues of the documents which are indicated as DOD adopted shall be the issue in the current DoDISS and the supplement thereto, if applicable.

AMERICAN NATIONAL STANDARDS INSTITUTE (ANSI)

ANSI B46.1 Surface Texture (Surface Roughness, Waviness, and Lay)

(Application for copies should be addressed to the American National Standards Institute, 1430 Broadway, New York, NY 10018.)

SHEET 5

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11

AMERICAN SOCIETY FOR TESTING AND MATERIALS (ASTM)

ASTM E 4 Load Verification of Testing Machines

ASTM E 74 Verification of Calibration Devices for Verifying Testing Machines

(Applications for copies should be addressed to the American Society for Testing and Materials, 1916 Race Street, Philadelphia, PA 19103.)

- 3. DEFINITIONS Not applicable.
- 4. GENERAL REQUIREMENTS
- 4.1 Test apparatus.
- 4.1.1 <u>Fixtures</u>. A typical test fixture arrangement for fatigue testing of fastener is illustrated in figure 1 and table I. Procedure to verify proper fatigue test machine and fixture alignment and cyclic-load application for the testing machine can be found in MIL-STD-1312 Appendix C.
- 4.1.2 <u>Static alignment</u>. Static alignment of each fatigue test machine shall be verified to ensure meeting the alignment requirements of MIL-STD-1312, Appendix C.
- 4.1.2.1 Spacer. A spacer conforming to figure l(c) may be used under the head of protruding head fasteners to provide clearance for the fastener head-to-shank fillet radius. Plain flat spacers may be used under the nut.
- 4.1.2.2 Test nuts. The fatigue test nuts shall conform to figure 2 and table II or figure 3 and table III. Threaded adapters may be used on shear fasteners through one-half inch diameter or tension fasteners too short to test with nuts on available fixtures. Locking devices are not permitted. Type A or B nuts may be used for testing bolts under 260 ksi. A fatigue test nut may be reused for not more than five tests, provided the nut meets the dimensional requirements and has not been damaged by a previous test and the same nut is used for testing the same lot of bolts. For referee tests, only unused nuts shall be used.
- 4.1.2.3 Test bolts. Test bolts or stude shall not be reused.
- 4.1.2.4 <u>Limitations</u>. Universal joints or spherical seats shall not be used anywhere on the test fixtures.

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11

TABLE I. List of items called out on figure 1.

Item	Description	Item	Description
1	Main mass	15	Impulse generator
2	Opposing mass	16	Driving magnet
3	Specimen (bolt)	17	Premagnetization (dc)
4	Preload spring	18	Main coils (ac)
5	Adjusting spindle	19	Tube amplifier
6	Spring supports	20	Phase adjustment
7	Dynamometer	21	Current waveforms
			With amplitude adjustment
8	Comparison bar	1	-Without amplitude adjust
9	Mirror bar	Α	Fixture preloading gear nut
10	Optical projector	В	Fixture stop plate
11	Dynamometer scale	С	Cup holding adapter
12	Diaphragm	D	Bolt loading cup
13	Photocell	1	
14	Photocell slide		

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11

TABLE II. List of dimensions for figure 2.

9 64

	A	В	H	J	K	P	Q	Y	
Thread	ref	min	+0.000	min	max	max	±0.010		
UNJF-3B			-0.030					Max	Min
01.11	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)
0.190-32	0.355	0,325	0.242	0.252	0.290	0.236	0.058	0.003	0.0005
0.250-28	0.481	0.441	0.349	0.380	0.434	0.296	0.082	0.003	0.0005
0.312-24	0.606	0.566	0.411	0.445	0.507	0.359	0.101	0.003	0.0005
0.375-24	0.740	0.700	0.497	0.509	0.580	0.421	0.122	0.003	0.0005
0.437-20	0.863	0.823	0.579	0.572	0.651	0.484	0.143	0.004	0.001
0.500-20	0.995	0.955	0.665	0.638	0.724	0.546	0.162	0.004	0.001
0.562-18	1.129	1.079	0.747	0.767	0.868	0.609	0.184	0.004	0.001
0.625-18	1.245	1.195	0.864	0.831	0.940	0.671	0.201	0.004	0.001
0.750-16	1.513	1.463	1.005	0.961	1.085	0.796	0.247	0.004	0.001
0.875-14	1.770	1.720	1.175	1.118	1.301	0.921	0.288	0.005	0.001
1-14NS	2.040	1.980	1.355	1.284	1.446	1.046	0.330	0.005	0.001
1.000-12	2.040	1.980	1.355	1.284	1.446	1.046	0.330	0.005	0.001
1.125-12	2.285	2.225	1.535	1.477	1.663	1.171	0.372	0.006	0.001
1.250-12	2.555	2.495	1.715	1.671	1.879	1.296	0.414	0.007	0.001
1.375-12	2.850	2.760	1.905	1.800	2.023	1.421	0.456	0.007	0.001
1.500-12	3.090	3.030	2.085	1.929	2.167	1.549	0.497	0.008	0.001

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11

TABLE III. List of dimensions for figure 3.

	A	J	Н	В	S	P		Y.
Thread	max	min	+0.000	min	ref	max		
UNJF-3B			-0.030				Max	Min
	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)	(in.)
0.190-32	0.434	0.380	0.151	0.367	0.057	0.210	0.003	0.0005
0.250-28	0.507	0.445	0.199	0.430	0.086	0.270	0.003	0.0005
0.312-24	0.580	0.509	0.253	0.492	0.120	0.332	0.003	0.0005
0.375-24	0.651	0.572	0.313	0.553	0.158	0.405	0.003	0.0005
0.437-20	0.797	0.703	0.373	0.679	0.189	0.467	0.004	0.001
0.500-20	0.868	0.767	0.439	0.741	0.220	0.530	0.004	0.001
0.562-18	1.013	0.896	0.509	0.865	0.262	0.597	0.004	0.001
0.625-18	1.085	0.961	0.584	0.928	0.307	0.660	0.004	0.001
0.750-16	1.229	1.089	0.750	1.052	0.505	0.781	0.004	0.001
0.875-14	1.446	1.284	0.925	1.239	0.630	0.906	0.005	0.001
1-14NS	1.663	1.477	1.115	1.427	0.776	1.031	0.005	0.001
1.000-12	1.663	1.477	1.115	1.427	0.776	1.031	0.005	0.001
1.125-12	1.879	1.671	1.309	1.614	0.712	1.156	0.006	0.001
1.250-12	2.096	1.865	1.528	1.801	0.832	1.281	0.007	0.001
1.375-12	2.312	2.058	1.761	1.988	0.960	1.406	0.007	0.001
1.500-12	2.528	2.253	2.010	2.176	1.097	1.531	0.008	0.001

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11

- 4.1.2.5 <u>Fixture alignment</u>. The fixtures used shall be capable of holding the fasteners in conformance with the alignment requirements specified in MIL-STD-1312, Appendix C.
- 4.1.2.6 Threaded adapters. The hardness of threaded adapters shall be as specified on figure 2.
- 4.2 Test specimens.
- 4.2.1 <u>Bolts</u>. Protruding head tensile bolts shall be tested using the fatigue test nuts as shown on figure 2 or the threaded adapter. Flush head bolts and protruding head shear bolts shall be tested using the fatigue test nuts (figures 2 and 3) or threaded adapters. Threaded adapters shall be replaced when they become damaged as determined by visual inspection and when they go out of tolerance. When adapters are used for referee tests, they shall be unused adapters. (Note: Fatigue tests are not applicable to parts with cross-drilled threads.)

For bolts having coarse or special threads, the fatigue test nut shall be as specified in the procurement specification.

The recommended minimum grip length for test bolts shall be twice the nominal diameter.

4.2.2 <u>Testing of nuts</u>. Nuts shall be tested on bolts as specified in the procurement specification. Nuts shall be assembled on the bolts with no less than two full threads exposed between the nut and the end of the bolt.

A sample of each lot of bolts or studs, used for testing nuts, shall first be tested in accordance with this test method using test nuts in figure 2 in order to determine their reliability.

- 4.2.3 <u>Nut test results</u>. Fracture in the threads of the bolt or nut and failure to support the test load shall constitute a valid test conclusion. Fracture of the bolt in any other location shall not be considered a valid test conclusion and shall be repeated using a new nut and bolt.
- 5. DETAIL REQUIREMENTS
- 5.1 Test procedures.
- 5.1.1 Bolts.
- 5.1.1.1 <u>Tension bolts</u>. Tension bolts shall be assembled so that a minimum of two and a maximum of three threads are exposed between the nut bearing face and bolt thread run out.
- 5.1.1.2 Shear bolts. Shear bolts shall be assembled with a minimum of one-half and a maximum of one thread exposed between the nut bearing face and the bolt thread run out.

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11

- 5.1.1.3 Flush head bolts. Flush head bolts shall be checked for uniform bearing between the bolt head and the loading cup in which it is to be tested. There shall be no contact in the area of the head-shank fillet radius.
- 5.2 Test conditions.
- 5.2.1 <u>Torque</u>. There shall be no tensile stress in the fasteners due to torque or other installation procedures.
- 5.2.2 <u>Load level</u>. The fatigue test load shall be as specified in the applicable procurement document or product specification.
- 5.2.3 <u>Load ratio</u>. Unless otherwise specified, the low load shall be one-tenth of the high-load level (R= 0.1).
- 5.2.4 <u>Test speed</u>. The test speed shall not exceed 12,500 cycles per minute. Excessive heat may be generated when testing high-strength fasteners at high frequencies. In this case the test speed shall be reduced to ensure that the maximum (test fasteners) temperature does not exceed 150°F.
- 6. NOTES
- 6.1 Test report. The test report shall contain the following:
 - a. Fastener description.
 - 1. Part number and fastener diameter.
 - 2. Lot identification.
 - 3 Material.
 - 4. Heat treatment.
 - 5. Grip length.
 - 6. Mating part.
 - 7. Measured fastener diameter.
 - b. Test Machine.
 - 1. Model and serial number.
 - 2. Calibration date.
 - c. Test load.
 - d. Test speed.
 - e. Test duration.
 - f. Type of failure.

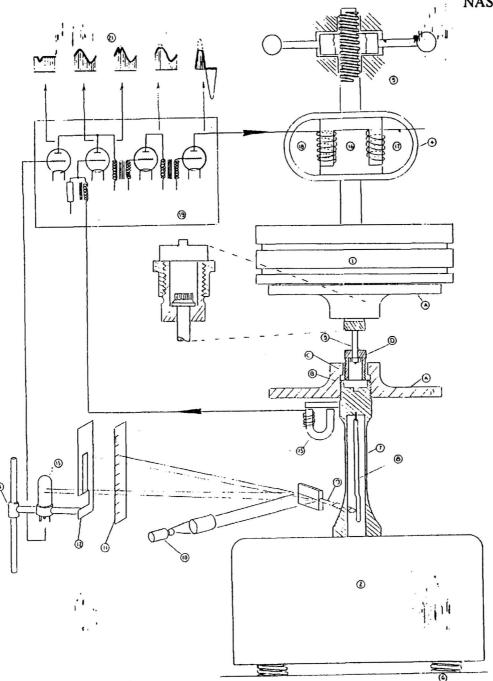
COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC.
ALL RIGHTS RESERVED

NASM1312-11

- g. Cause and duration of any interruption during test.
- h. Installation procedure.
- i. Results of all inspections.

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC ALL RIGHTS RESERVED

NASM1312-11



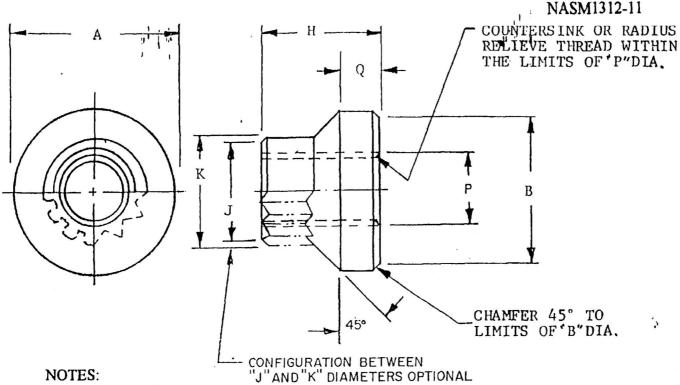
NOTE:

1. The list of numbered and lettered items appear in table I.

FIGURE 1. Typical test fixture arrangement.

SHEET 13

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC
ALL RIGHTS RESERVED



1. Material: Alloy steel.

- 2. Hardness: Type A HRC 27-32 for testing bolts rated below 260 ksi
 Type B HRC 36-42 for testing bolts rated 260 ksi and higher
- 3. Heat treat per MIL-H-6875.
- 4. Surface texture: 125 microinches per ANSI B46.1.
- 5. Protective coating: Rust preventive oil (MIL-C-15074) to be removed before testing.
- 6. Residual traces of cadmium permitted on nuts retapped to remove locking device.
- 7. Magnetic particle inspect per MIL-I-6868; omit identification.
- 8. Threads per MIL-S-8879.
- 9. Concentricity: 'All diameters to be concentric within 0.005 FIM.
- 10. Squareness: Bearing surface to thread P.D. within 0.003 FIM.
- 11. Bearing surface to be concave within "Y" FIM.
- 12. Dimensions in inches unless otherwise specified (see table II).
- 13. Break all sharp edges.

Tolerances: Three decimals ±0.010 unless otherwise noted; Angular ±2°.

FIGURE 2. Nut, fatigue test, room temperature, for tension bolts.

SHEET 14

©COPYRIGHT 1997 AEROSPACE INDUSTRIES ASSOCIATION OF AMERICA, INC.
ALL RIGHTS RESERVED

CONFIGURATION BETWEEN
A"AND "J"DIAS. IS OPTIONAL

UPPER PORTION OF
NUT SHALL BE TAPERED

BELOW"J"DIA.

NOTES:

- 1. Material: Alloy steel.
- 2. Hardness: HRC 36-42.
- 3. Heat treat per MIL-H-6875.
- 4. Surface texture: 125 microinches per ANSI B46.1.
- 5. Protective Coating: Rust preventive oil (MIL-C-15074) to be removed before testing.
- 6. Residual traces of cadmium permitted on nuts retapped to remove locking device.
- 7. Magnetic particle inspect per MIL-I-6868; omit identification.
- 8. Threads per MIL-S-8879.
- 9. Concentricity: 'All diameters to be concentric within 0.005 FIM.
- 10. Squareness: Bearing surface to thread P.D. within 0,003 FIM.
- 11. Bearing surface to be concave within "Y" FIM.
- 12. Dimensions in inches unless otherwise specified (see table III).
- 13. Break all sharp edges.
- 14. Tolerances: Three decimals ± 0.010 unless otherwise noted; Angular $\pm 2^{\circ}$.

FIGURE 3. Nut fatigue test, room temperature, for shear bolts.

SHEET 15